

IN THE ABSTRACT:

Please replace the Abstract with the following:

A method facilitates assembling and apparatus for a rotor assembly for gas turbine engine are provided. The method includes providing a first rotor blade including an airfoil, a platform, a shank, an internal cavity, and a dovetail is provided, wherein the airfoil extends radially outward from the platform, which includes a radially outer surface and a radially inner surface, the shank extends radially inward from the platform, and the dovetail extends from the shank, such that the internal cavity is defined by the airfoil, the platform, the shank, and the dovetail. The method also includes coupling the first rotor blade is coupled to a rotor shaft such that during engine operation, cooling air is channeled from the cavity through an impingement cooling circuit for impingement cooling the first rotor blade platform radially inner surface, and coupling a second rotor blade is coupled to the rotor shaft such that a platform gap is defined between the first and second rotor blade platforms.